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Collette

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(54) **FUEL LINE AIR TRAP FOR AN UNMANNED AERIAL VEHICLE**

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B64D 33/00 (2006.01)

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(58) **Field of Classification Search** .... 244/135 R, 244/135 A, 53 R; 137/177, 179, 197, 590, 137/592; 138/30, 39, 44; 123/516, 198 E; 220/601

See application file for complete search history.

(56) **References Cited**

U.S. PATENT DOCUMENTS

2,326,175	A	8/1943	Samiran	
2,383,369	A	8/1945	Curtis	
2,952,214	A	9/1960	Adams	
3,212,521	A *	10/1965	Jean	137/494
4,212,600	A	7/1980	Otto et al.	
4,231,719	A	11/1980	Ringwald et al.	
4,309,155	A	1/1982	Heinz et al.	
5,146,901	A	9/1992	Jones	
5,150,857	A	9/1992	Moffitt et al.	
5,152,478	A	10/1992	Cycon et al.	
5,295,643	A	3/1994	Ebbert et al.	
5,355,860	A	10/1994	Ekstam	
5,372,116	A *	12/1994	Davis	123/516
5,575,438	A	11/1996	McGonigle et al.	
5,695,153	A	12/1997	Britton et al.	
5,746,184	A	5/1998	Ekstam	
6,450,445	B1	9/2002	Moller	
6,604,706	B1	8/2003	Bostan	
6,691,949	B2	2/2004	Plump et al.	
6,721,646	B2	4/2004	Carroll	
7,032,861	B2	4/2006	Sanders, Jr. et al.	
7,044,422	B2	5/2006	Bostan	
7,249,732	B2	7/2007	Sanders, Jr. et al.	
2003/0234002	A1	12/2003	Rivera	

(Continued)

FOREIGN PATENT DOCUMENTS

EP 1767453 A1 3/2007

(Continued)

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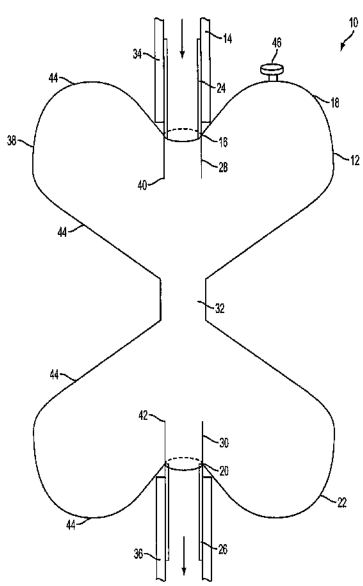
(57) **ABSTRACT**

A fuel line air trap for an unmanned aerial vehicle has a vessel in-line with a fuel line, fuel line connectors, and fuel inlet and outlet stems. The vessel contains an inlet at a distal end and an outlet at a proximal end. The fuel line connectors are attached to the vessel at the inlet and at the outlet to connect the fuel line to the vessel. The fuel inlet stem attaches to the vessel at the inlet and a fuel outlet stem attaches to the vessel at the outlet. Both the fuel inlet stem and fuel outlet stem protrude into the vessel a predetermined distance such that a gap exists between them. As air bubbles enter the gap in the fuel line air trap, they separate from the flow of fuel and migrate to the exterior walls of the vessel. The air bubbles are purged during remain trapped refueling.

20 Claims, 4 Drawing Sheets



U.S. PATENT DOCUMENTS

2004/0094662	A1	5/2004	Sanders, Jr. et al.
2004/0129828	A1	7/2004	Bostan
2005/0082421	A1	4/2005	Perlo et al.
2006/0049304	A1	3/2006	Sanders, Jr. et al.
2006/0192047	A1	8/2006	Goossen
2006/0231675	A1	10/2006	Bostan
2007/0034738	A1	2/2007	Sanders, Jr. et al.
2007/0051848	A1	3/2007	Mantych et al.
2007/0193650	A1	8/2007	Annati
2007/0221790	A1	9/2007	Goossen et al.
2007/0228214	A1	10/2007	Horak
2007/0244608	A1	10/2007	Rath et al.
2007/0262195	A1	11/2007	Bulaga et al.
2007/0295298	A1	12/2007	Mark

FOREIGN PATENT DOCUMENTS

EP	1767453 B1	6/2008
WO	0015497	3/2000
WO	2004002821	1/2004

* cited by examiner

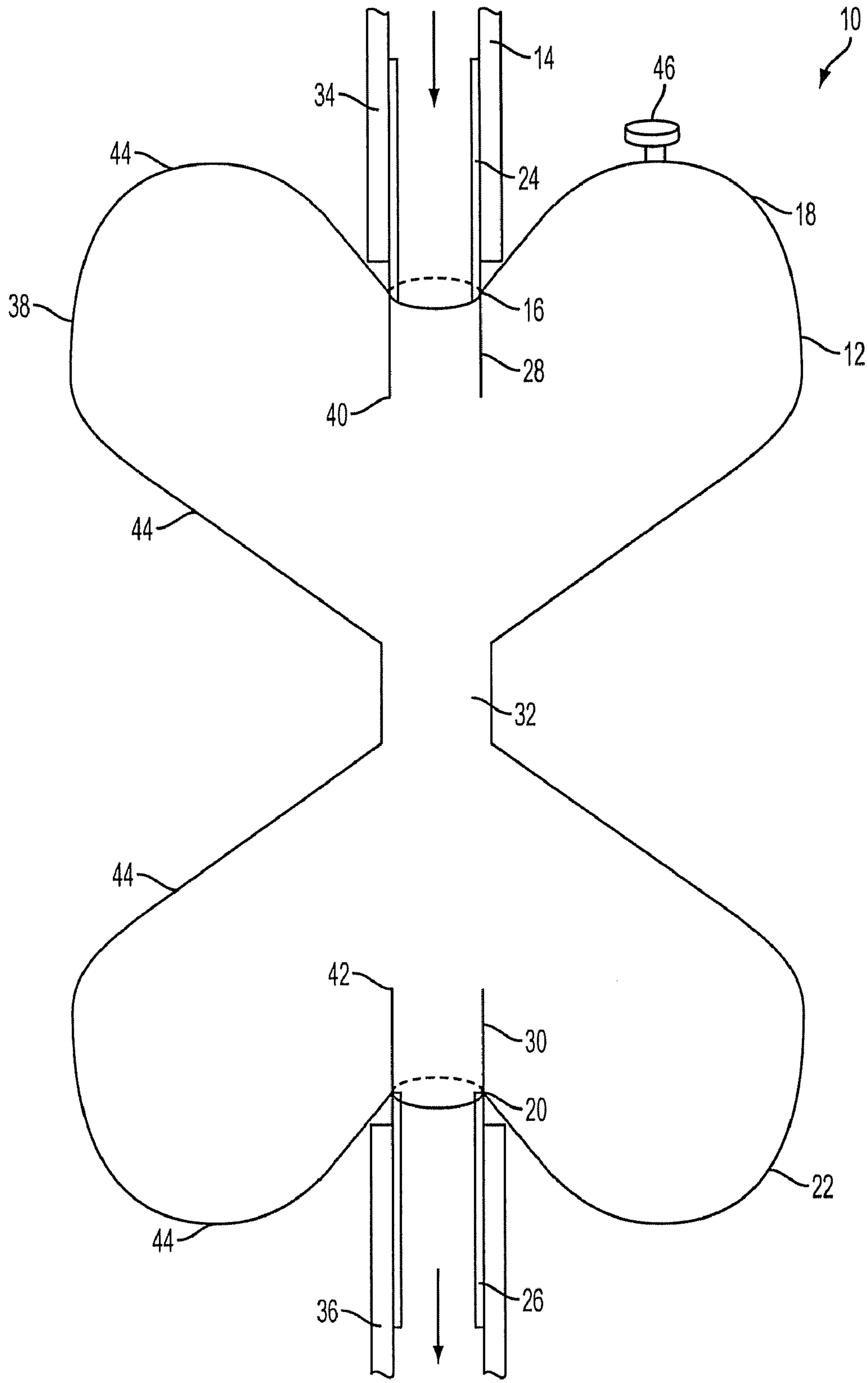


FIG. 1

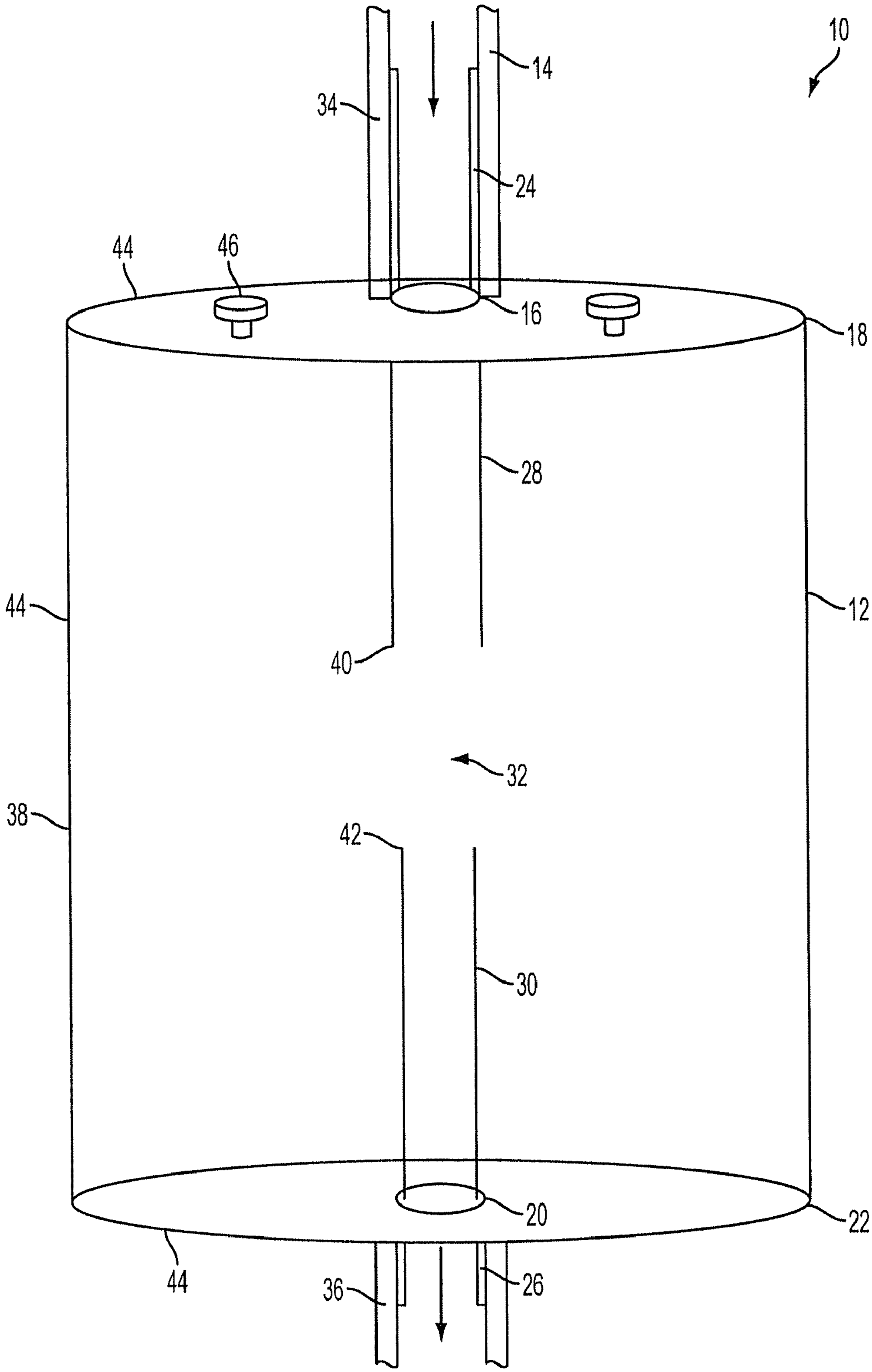
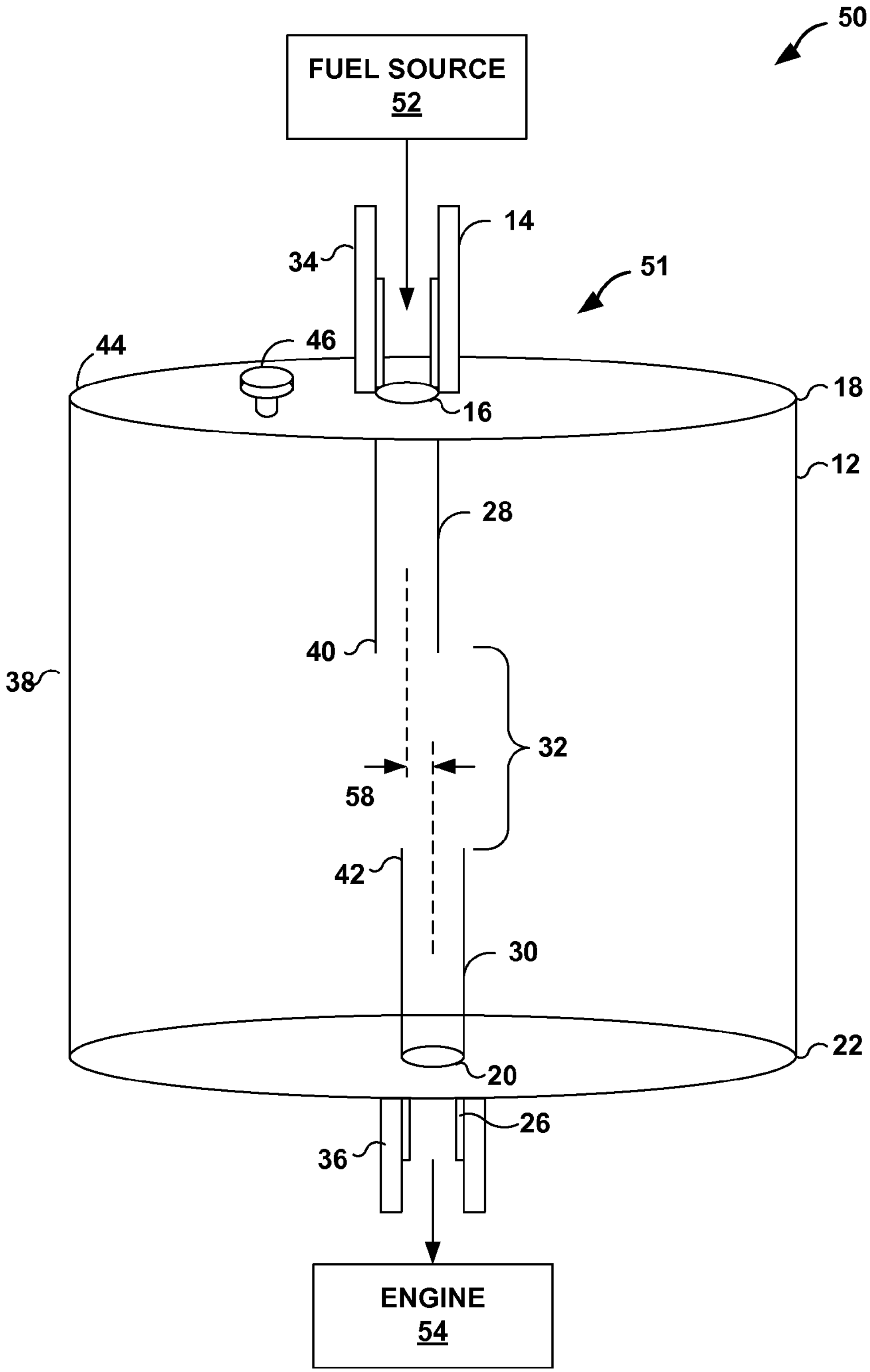


FIG. 2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ENGINE
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FIG. 3

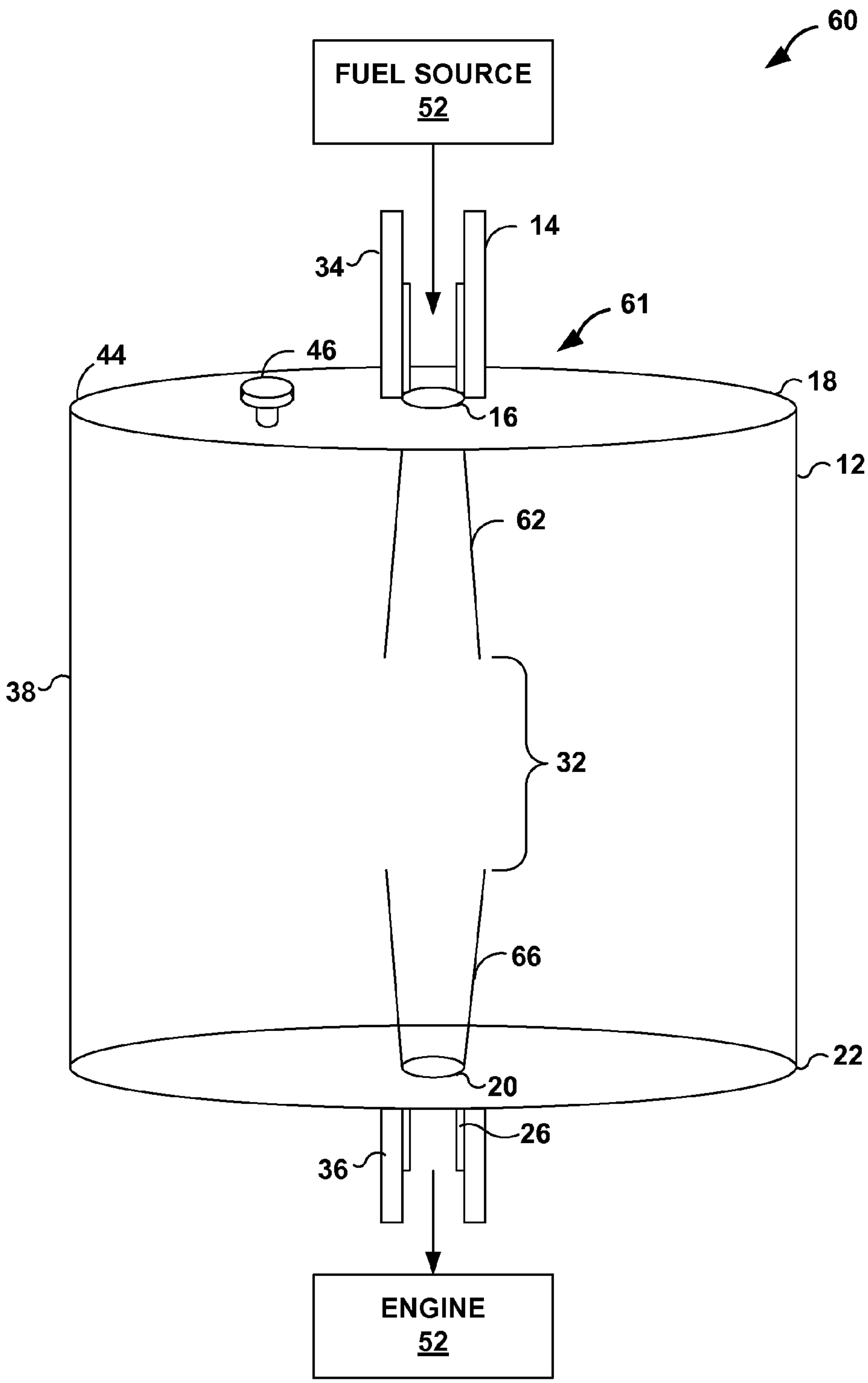
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FIG. 4

FUEL LINE AIR TRAP FOR AN UNMANNED AERIAL VEHICLE

GOVERNMENT RIGHTS

The United States Government may have acquired certain rights in this invention pursuant to Government Contract No. W56HZV-05-C-0724 with the U.S. Army (TACOM).

BACKGROUND OF THE INVENTION

Unmanned aerial vehicle (UAV) or micro-air vehicle (MAV) applications that rely on a vacuum from the engine to draw the fuel, as opposed to having a pump in the gas reservoir, may generate air bubbles in the fuel line. In addition, UAVs often encounter turbulence and adverse flying conditions that may introduce air into the fuel line such as when the vehicle pitches or engages in forward flight on a low fuel reserve. Similarly, fuel drawn by a vacuum in an operating environment like the desert where there are elevated temperatures is particularly susceptible to vapor lock caused by air in the fuel line. As such, any of these situations may result in the loss of an air vehicle by starving the engine of fuel.

One apparatus employed to regulate the proper amount of fuel fed to the engine is the carburetor. Carburetors typically contain a float chamber that holds a quantity of fuel for immediate use. This reservoir is constantly replenished with fuel supplied by a fuel pump. The correct fuel level in the chamber is maintained by a float that cooperates with the opening of an inlet valve. As the fuel level is depleted by the engine, the float drops accordingly, opening the inlet valve and admitting fuel. As the fuel level rises, the float rises and seals the inlet valve. Usually, ventilation tubes allow air to escape from the chamber as it fills with fuel or air to enter as the chamber empties, maintaining atmospheric pressure within the float chamber.

However, where the engine may be operated in various orientations relative to the ground, such as in a UAV, the float chamber of a carburetor is rendered useless due to its dependence on gravity. In addition, there is a significant weight associated with the fuel pump and float chamber, which is a key reason for not utilizing them in UAVs. To solve this problem, a flexible diaphragm may be utilized to form a wall of the fuel chamber so that as fuel is drawn into the engine the diaphragm is forced inward by ambient air pressure. The diaphragm is connected to a needle valve and as the diaphragm moves into the chamber the needle valve is forced open to admit more fuel. As fuel enters the chamber, the diaphragm expands outward, closing the needle valve. A balanced state is reached which creates a steady fuel level that remains constant in any orientation. This modified carburetor lacks an air venting mechanism and thus does not resolve the vapor lock issue presented by UAVs.

SUMMARY OF THE INVENTION

The discovery presented herein outlines an apparatus for trapping and removing air bubbles from a fuel line undergoing vacuum pressure generated by an engine, where the apparatus functions in any orientation.

Thus, in a first aspect, the present invention provides a fuel line air trap for an unmanned aerial vehicle comprising: (a) a vessel in-line with a fuel line, wherein the vessel contains an inlet at a distal end and an outlet at a proximal end, (b) a distal fuel line connector and a proximal fuel line connector attached to the vessel at the inlet and at the outlet, respectively, for attaching the fuel line to the vessel, and (c) a fuel inlet stem attached to the vessel at the inlet and a fuel outlet stem attached to the vessel at the outlet, wherein both the fuel inlet stem and fuel outlet stem protrude into the vessel a predetermined distance such that a gap exists between the fuel inlet stem and fuel outlet stem.

The present invention also provides a fuel line air trap for an unmanned aerial vehicle comprising: (a) a vessel in-line with a fuel line, wherein the vessel is positioned sufficiently close to an engine such that any vaporization that occurs in the fuel line between the air trap and the engine will not generate a vapor lock in the engine, wherein the vessel contains an inlet at a distal end and an outlet at a proximal end, wherein the inlet and the outlet are concentric to each other, (b) a distal fuel line connector and a proximal fuel line connector attached to the vessel at the inlet and at the outlet, respectively, for attaching the fuel line to the vessel, and (c) a fuel inlet stem connected to the vessel at the inlet and a fuel outlet stem connected to the vessel at the outlet such that the fuel inlet stem and fuel outlet stem both protrude into the vessel a predetermined distance such that a gap exists between the fuel inlet stem and fuel outlet stem, wherein the volume contained by the vessel between the vessel's distal end and a cross-section of the vessel in a plane substantially perpendicular to the fuel line taken where the fuel inlet stem terminates in the gap and the volume contained by the vessel between the vessel's proximal end and a cross-section of the vessel in a plane substantially perpendicular to the fuel line taken where the fuel outlet stem terminates in the gap are each at least substantially equal to the total volume of air that will be generated before refueling is necessary.

The present invention further provides a fuel line air trap for an unmanned aerial vehicle comprising: (a) a vessel in-line with a fuel line, wherein the vessel is in the form of a cylinder, wherein the vessel is positioned 12 cm or less from an engine to minimize fuel vaporization in the fuel line between the vessel and the engine, wherein the vessel contains an inlet at a distal end and an outlet at a proximal end, wherein the inlet and the outlet are concentric to each other, (b) a distal fuel line connector and a proximal fuel line connector attached to the vessel at the inlet and at the outlet, respectively, for attaching the fuel line to the vessel, and (c) a fuel inlet stem attached to the vessel at the inlet and a fuel outlet stem attached to the vessel at the outlet, wherein both the fuel inlet stem and fuel outlet stem protrude into the vessel a predetermined distance such that a gap exists between the fuel inlet stem and fuel outlet stem.

BRIEF DESCRIPTION OF THE DRAWINGS

FIG. **1** is a cross-sectional view of one embodiment of the air trap.

FIG. **2** is cross-sectional view of another embodiment of the air trap.

FIG. **3** is a cross-sectional view of one embodiment of a system including an air trap comprising an inlet and an outlet eccentric to one another.

FIG. **4** is a cross-sectional view of one embodiment of a system including an air trap comprising a tapered fuel inlet stem and a tapered fuel outlet stem.

DETAILED DESCRIPTION OF THE PREFERRED EMBODIMENT

In a first aspect, the present invention provides a fuel line air trap **10** for an unmanned aerial vehicle comprising: (a) a vessel **12** in-line with a fuel line **14**, wherein the vessel **12** contains an inlet **16** at a distal end **18** and an outlet **20** at a proximal end **22**, (b) a distal fuel line connector **24** and a proximal fuel line connector **26** attached to the vessel **12** at the inlet **16** and at the outlet **20**, respectively, for attaching the fuel line **14** to the vessel **12**, and (c) a fuel inlet stem **28** attached to the vessel **12** at the inlet **16** and a fuel outlet stem **30** attached to the vessel **12** at the outlet **20**, wherein both the fuel inlet stem **28** and fuel outlet stem **30** protrude into the vessel **12** a predetermined distance such that a gap **32** exists between the fuel inlet stem **28** and fuel outlet stem **30**.

As used herein, the fuel line **14** is divided into two sections **34**, **36** such that the first section **34** is connected between a fuel source (shown in FIG. **3**) and the vessel **12** and the second section **36** is connected between the vessel **12** and the engine (shown in FIG. **3**). The centerlines of the two sections **34**, **36** of the fuel line **14** may either lie on substantially the same axis or substantially parallel axes.

As used herein, the vessel **12** consists of a body **38**, an inlet **16**, an outlet **20**, a fuel inlet stem **28**, a fuel outlet stem **30**, a distal fuel line connector **24**, and a proximal fuel line connector **26**. The vessel body **38** is in-line with the fuel line **14** such that the end of the vessel **12** closest to the fuel source is the distal end **18**, whereas the end closest to the engine is the proximal end **22**. The vessel body **38** may take many forms, including for example an hourglass (see FIG. **1**), cylinder (see FIG. **2**), box, sphere, etc. The vessel **12** is preferably made of any material not affected by gas or diesel fuels, for example polypropylene plastic. The vessel **12** is also preferably transparent or translucent to assist in determining whether air has been purged from the fuel system during refueling.

The distal end **18** of the vessel **12** contains an inlet **16** and the proximal end **22** contains an outlet **20**. The distal and proximal fuel line connectors **24**, **26** are located at the vessel inlet **16** and vessel outlet **20**, respectively. These connectors **24**, **26** may be molded as part of the vessel body such that they protrude from the exterior of the vessel **12** in the form of stems. In this stem configuration, the fuel line connectors **24**, **26** have an outer diameter that allows each connector to fit snugly within the fuel line **14** and be secured with any type of clamp (not shown) generally known in the art. Alternatively, the external surface of the stem fuel line connectors **24**, **26** may be threaded to connect to a fuel line **14** with corresponding threads. The fuel line connectors **24**, **26** may similarly be recessed in the vessel body and threaded to receive a fuel line **14** with corresponding threads.

The fuel inlet stem **28** and fuel outlet stem **30** are also located at the vessel's inlet **16** and outlet **20**. These fuel inlet and outlet stems **28**, **30** may be molded as part of the vessel body such that they protrude into the vessel **12** and are located between the fuel line connectors **24**, **26**. The fuel stems **28**, **30** extend a predetermined distance such that a gap **32** exists between the fuel inlet and outlet stems **28**, **30**. The gap **32** is small enough for the fuel outlet stem **30** to draw from the fuel in the gap **32**, rather than from adjacent accumulated air, and wide enough that the velocity of the fuel entering the vessel **12** will not carry an air bubble across the gap **32** into the fuel outlet stem **30**. This gap distance is preferably 12 cm. The predetermined distance each stem **28**, **30** extends is preferably of a length such that the fuel inlet and outlet stems **28**, **30** do not overlap, for example, in the case where the stems **28**, **30** are not concentrically aligned.

In operation, the UAV or MAV is purged of air and fully fueled. In this state the fuel source, fuel line **14**, and air trap **10** are completely filled with fuel. As the UAV or MAV consumes fuel during the course of a flight, air bubbles are generated in the system due to operating conditions, such as high temperatures, turbulence, a low fuel reserve, etc. These air bubbles are carried through the fuel line **14** to the air trap **10**. The fuel enters the vessel **12** through the fuel inlet stem **28**. When the air bubbles reach the gap **32**, the velocity of the fuel slows to a degree that allows the air bubbles to separate from the fuel flow and float to the exterior wall **44**, **18**, **22** of the air trap's vessel **12**, as opposed to being pulled into the fuel outlet stem **30**. If the UAV is in a standard forward flight orientation, the air bubbles will float to the top, which in this aspect is the distal end **18**. If the UAV is maneuvering in various orientations, the accumulated air pocket in the air trap **10** will travel along the walls **44** of the vessel **12** to the highest point relative to the ground, and any entering air bubbles will travel through the fuel to this air pocket. Since the accumulated air travels along the perimeter of the vessel **12**, the fuel outlet stem **30** continues to draw only fuel from the gap **32**.

In one embodiment, the vessel **12** is positioned sufficiently close to an engine such that any vaporization that occurs in the fuel line **14** between the air trap **10** and the engine will not generate a vapor lock in the engine. The required distance from the air trap **10** to the engine is primarily a function of the vacuum generated by the engine and the environmental operating temperature. Air bubbles are generated faster in the presence of a stronger vacuum and higher temperature. For instance, in a desert environment where temperatures reach upwards of 100 degrees, the air trap is preferably 5 to 12 cm from the engine. Likewise, when the vacuum generated is lower and/or the environmental temperature is lower, the air trap can be greater than 12 cm away from the engine. Another factor to consider is that as the vacuum in the fuel line **14** increases due to the amount of unconsumed fuel decreasing, the fuel tends to vaporize more easily as it is drawn through the fuel line **14**. So an air trap **10** at a distance greater than 12 cm may be functional during the initial portion of the flight but could fail under an increasing vacuum in later flight. This makes clear that the closeness of the air trap **10** to the engine is dependent on the application and conditions in which the particular UAV or MAV is designed to fly under. Thus, by placing the air trap **10** sufficiently close to the engine any vaporization that occurs in the fuel line **14** between the air trap **10** and the engine will be negligible.

In one embodiment, the volume contained by the vessel **12** between the vessel's distal end **18** and the point **40** at which the fuel inlet stem **28** terminates in the gap **32** and the volume contained by the vessel **12** between the vessel's proximal end **22** and the point **42** at which the fuel outlet stem terminates **30** in the gap **32** are each at least equal to the total volume of air that will be generated before refueling is necessary. By configuring the vessel's volume in this manner, the accumulated air will remain adjacent to the vessel wall and the inlet or outlet fuel stems **28**, **30**, depending on the orientation of the UAV in flight, and will avoid being drawn into the fuel flow in the gap **32**. If the accumulated air were to exceed the volume at the distal end **18** near the fuel inlet stem **28**, the air pocket would in effect shorten the gap **32** and the force of the fuel flow from the fuel inlet stem **28** could introduce the air into the fuel outlet stem **30**. Similarly, if the accumulated air pocket were to travel along the vessel walls **44** to the proximal end **22** and exceed the volume of the vessel adjacent to the fuel outlet stem **30**, the air would be drawn directly into the fuel outlet stem **30**.

FIG. **3** is a cross-sectional view of one embodiment of a system **50** including an air trap **51** comprising an inlet **16** and an outlet **20** eccentric to one another. The system **50** includes a fuel source **52** coupled to the air trap **51** at a distal end and an engine **54** coupled to the air trap **51** at a proximal end. In one embodiment, the inlet **16** is offset from the center of the vessel's distal end **18** and the outlet **20** is located at the center of the vessel's proximal end **22** such that the inlet **16** and outlet **20** are eccentric. The inlet **16** and the outlet **20** are offset by a distance **58** in the example shown in FIG. **3**. By offsetting the inlet **16** and outlet **20**, the fuel inlet and outlet stems **28**, **30** are likewise offset. This eccentric configuration reduces the velocity of the fuel across the gap **32**, which allows air bubbles to separate more easily from the flow of the fuel and travel to the accumulated air pocket. Though both the fuel inlet and outlet stem **28**, **30** could be offset from the center of each respective end **18**, **22** of the vessel **12**, offsetting the fuel inlet stem **28** is preferred. By retaining the fuel outlet stem **30** in the center, the vacuum draw is kept away from the sidewalls **44** of the vessel **12** and therefore away from the accumulated air pocket.

In one embodiment, the inlet **16** and the outlet **20** are substantially concentric to each other. This configuration is preferred over the eccentric alignment of the inlet **16** and outlet **20**. In the eccentric alignment there is a possibility that as an air bubble enters the vessel **12** the UAV may alter its orientation such that the fuel outlet stem is directly in the air bubble's path to the air pocket. Though this situation would be unlikely, aligning the fuel inlet and outlet stems **28**, **30** concentrically avoids this situation altogether. As an air bubble enters the vessel **12** in a concentric configuration, the fuel outlet stem **30** would only reside in the path of an air bubble if the UAV were completely upside down and flying straight towards the ground. However, this upside down flight orientation is the least likely maneuver to occur during a UAV flight.

FIG. **4** is a cross-sectional view of one embodiment of a system **60** including an air trap **61** having a tapered fuel inlet stem **62** and a tapered fuel outlet stem **66**. In one embodiment, the diameters of the fuel inlet stem and fuel outlet stem **62**, **66** are tapered to be larger near the gap than the diameters at the inlet **16** and outlet **20**, respectively. By tapering the diameter of the fuel inlet and outlet stems **62**, **66** in this manner, the velocity of the fuel is slowed. A slower velocity across the gap **32** is advantageous because it allows the air more time to separate from the fuel flow and accumulate in the air trap **10**.

In one embodiment, the diameter of the vessel **12** at the midpoint of the gap **32** between the fuel inlet stem **62** and the fuel outlet stem **66** is larger than the diameter of the fuel inlet stem **62** at the inlet **16** and the diameter of the fuel outlet stem at the outlet **20**. This configuration again slows the velocity of the fuel through the air trap **10** and allows the air more time to separate from the fuel flow.

In one embodiment, the fuel line air trap **10** further comprises one or more release valves **46** connected to the vessel **12** to purge trapped air. These one or more release valves **46** are manually opened during refuel to allow air to be purged from the system.

In one embodiment, the one or more release valves **46** are located at the highest point of the vessel **12** relative to the ground when the UAV is at rest. In this aspect of the invention, the distal end **18** of the vessel **12** represents the highest point of the air trap **10** when the UAV is at rest. Thus, as fuel enters the air trap **10** during refuel, air will be purged from the release valves **46**. If it were otherwise, fuel would spill from the valves **46** before all the air in the trap **10** was purged.

In one embodiment, the center of the gap **32** is located substantially midway between the distal and proximal ends **18**, **22**.

In one embodiment, the vessel **12** is substantially symmetrical on either side of the gap **32**.

In one embodiment, the vessel **12** is in the form of a cylinder (see FIG. **2**).

In one embodiment, the vessel **12** is in the form of an hourglass (see FIG. **1**).

In a second aspect, the present invention may take the form of a fuel line air trap **10** for an unmanned aerial vehicle comprising: (a) a vessel **12** in-line with a fuel line **14**, wherein the vessel **12** is positioned sufficiently close to an engine such that any vaporization that occurs in the fuel line **14** between the air trap **10** and the engine will not generate a vapor lock in the engine, wherein the vessel **12** contains an inlet **16** at a distal end **18** and an outlet **20** at a proximal end **22**, wherein the inlet **16** and the outlet **20** are concentric to each other, (b) a distal fuel line connector **24** and a proximal fuel line connector **26** attached to the vessel **12** at the inlet **16** and at the outlet **20**, respectively, for attaching the fuel line **14** to the vessel **12**, and (c) a fuel inlet stem **28** is connected to the vessel **12** at the inlet **16** and a fuel outlet stem **30** is connected to the vessel **12** at the outlet **20** such that the fuel inlet stem **28** and fuel outlet stem **30** both protrude into the vessel **12** a predetermined distance such that a gap **32** exists between the fuel inlet stem **28** and fuel outlet stem **30**, wherein the volume contained by the vessel **12** between the vessel's distal end **18** and the point **40** at which the fuel inlet stem terminates in the gap **32** and the volume contained by the vessel **12** between the vessel's proximal end **22** and the point **42** at which the fuel outlet stem terminates in the gap **32** are each at least substantially equal to the total volume of air that will be generated before refueling is necessary.

In one embodiment, the center of the gap **32** is located substantially midway between the distal and proximal ends **18**, **22**.

In another embodiment, the fuel line air trap **10** further comprises one or more release valves **46** connected to the vessel **12** to purge trapped air.

In a further embodiment, the one or more release valves **46** are located at the highest point of the vessel relative to the ground when the UAV is at rest.

In a third aspect, the present invention may take the form of a fuel line air trap for an unmanned aerial vehicle comprising: (a) a vessel **12** in-line with a fuel line **14**, wherein the vessel **12** is in the form of a cylinder (see FIG. **2**), wherein the vessel **12** is positioned 12 cm or less from an engine to minimize fuel vaporization in the fuel line **14** between the vessel **12** and the engine, wherein the vessel **12** contains an inlet **16** at a distal end **18** and an outlet **20** at a proximal end **22**, wherein the inlet **16** and the outlet **20** are concentric to each other, (b) a distal fuel line connector **24** and a proximal fuel line connector **26** attached to the vessel **12** at the inlet **16** and at the outlet **20**, respectively, for attaching the fuel line **14** to the vessel **12**, and (c) a fuel inlet stem **28** attached to the vessel **12** at the inlet **16** and a fuel outlet stem **30** attached to the vessel **12** at the outlet **20**, wherein both the fuel inlet stem **28** and fuel outlet stem **30** protrude into the vessel **12** a predetermined distance such that a gap **32** exists between the fuel inlet stem **28** and fuel outlet stem **30**.

In one embodiment, the diameters of the fuel inlet stem **28** and fuel outlet stem **30** are tapered to be larger near the gap than the diameters at the inlet **16** and outlet **20**, respectively.

In another embodiment, the fuel line air trap **10** further comprises one or more release valves **46** connected to the vessel **12** to purge trapped air.

The invention claimed is:

1. A system for an unmanned aerial vehicle, the system comprising:

a fuel line air trap, comprising:

a vessel in-line with a fuel line, wherein the vessel comprises an inlet at a distal end and an outlet at a proximal end;

a distal fuel line connector and a proximal fuel line connector attached to the vessel at the inlet and at the outlet, respectively, and configured to attach the fuel line to the vessel; and

a fuel inlet stem attached to the vessel at the inlet and a fuel outlet stem attached to the vessel at the outlet, wherein both the fuel inlet stem and fuel outlet stem protrude into the vessel a predetermined distance such that a gap exists between the fuel inlet stem and fuel outlet stem, wherein at least a portion of the gap is in-line with the fuel line.

2. The system of claim **1**, further comprising:

an engine, wherein the vessel is positioned sufficiently close to the engine such that any vaporization that occurs in the fuel line between the air trap and the engine will not generate a vapor lock in the engine.

3. The system of claim **1**, wherein a volume contained by the vessel between the vessel's distal end and a point at which the fuel inlet stem terminates in the gap and a volume contained by the vessel between the vessel's proximal end and a point at which the fuel outlet stem terminates in the gap are each at least substantially equal to a total volume of air that will be generated before refueling is necessary.

4. The system of claim **1**, wherein the inlet is offset from a center of the vessel's distal end and the outlet is located at a center of the vessel's proximal end such that the inlet and outlet are eccentric.

5. The system of claim **1**, wherein the inlet and the outlet are substantially concentric to each other.

6. The system of claim **1**, wherein the fuel inlet stem is tapered such that a diameter of the fuel inlet stem near the gap is larger than a diameter of the fuel inlet stem near the inlet, and the fuel outlet stem is tapered such that a diameter of the fuel outlet stem near the gap is larger than a diameter of the fuel outlet stem near the outlet.

7. The system of claim **1**, wherein a diameter of the vessel at a midpoint of the gap between the fuel inlet stem and the fuel outlet stem is larger than a diameter of the fuel inlet stem at the inlet and a diameter of the fuel outlet stem at the outlet.

8. The system of claim **1**, further comprising one or more release valves connected to the vessel to purge trapped air.

9. The system of claim **8**, wherein the one or more release valves are located at a highest point of the vessel relative to a ground level when the unmanned aerial vehicle (UAV) is at rest.

10. The system of claim **1**, wherein a center of the gap is located substantially midway between the distal and proximal ends.

11. The system of claim **1**, wherein the vessel is substantially symmetrical on either side of the gap.

12. The system of claim **11**, wherein the vessel is in the form of a cylinder.

13. The system of claim **11**, wherein the vessel is in the form of an hourglass.

14. A fuel line air trap for an unmanned aerial vehicle comprising:

a vessel in-line with a fuel line, wherein the vessel comprises an inlet at a distal end and an outlet at a proximal end, wherein the inlet and the outlet are concentric to each other;

a distal fuel line connector and a proximal fuel line connector attached to the vessel at the inlet and at the outlet, respectively, and configured to attach the fuel line to the vessel; and

a fuel inlet stem connected to the vessel at the inlet and a fuel outlet stem connected to the vessel at the outlet such that the fuel inlet stem and fuel outlet stem both protrude into the vessel a predetermined distance such that a gap exists between the fuel inlet stem and fuel outlet stem at least partially in-line with the fuel line, wherein a volume contained by the vessel between the vessel's distal end and a point at which the fuel inlet stem terminates in the gap and a volume contained by the vessel between the vessel's proximal end and a point at which the fuel outlet stem terminates in the gap are each at least substantially equal to a total volume of air that will be generated before refueling is necessary.

15. The fuel line air trap of claim **14**, wherein the center of the gap is located substantially midway between the distal and proximal ends.

16. The fuel line air trap of claim **14**, further comprising one or more release valves connected to the vessel to purge trapped air.

17. The fuel line air trap of claim **16**, wherein the one or more release valves are located at a highest point of the vessel relative to a ground level when the unmanned aerial vehicle (UAV) is at rest.

18. A system for an unmanned aerial vehicle comprising:

an engine; and

a fuel line air trap, comprising:

a vessel in-line with a fuel line, wherein the vessel is in the form of a cylinder, wherein the vessel is positioned approximately 12 centimeters (cm) or less from the engine to minimize fuel vaporization in the fuel line between the vessel and the engine, wherein the vessel comprises an inlet at a distal end and an outlet at a proximal end, wherein the inlet and the outlet are concentric to each other;

a distal fuel line connector and a proximal fuel line connector attached to the vessel at the inlet and at the outlet, respectively, for attaching the fuel line to the vessel; and

a fuel inlet stem attached to the vessel at the inlet and a fuel outlet stem attached to the vessel at the outlet, wherein both the fuel inlet stem and fuel outlet stem protrude into the vessel a predetermined distance such that a gap exists in-line with the fuel line between the fuel inlet stem and fuel outlet stem.

19. The system of claim **18**, wherein the fuel inlet stem is tapered such that a diameter of the fuel inlet stem near the gap is larger than a diameter of the fuel inlet stem near the inlet, and the fuel outlet stem is tapered such that a diameter of the fuel outlet stem near the gap is larger than a diameter of the fuel outlet stem near the outlet.

20. The system of claim **18**, further comprising one or more release valves connected to the vessel to purge trapped air.

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